



NASA Rotor 37 CFD Code Validation

Glenn-HT Code

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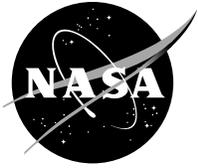
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Abstract

In order to advance the goals of NASA aeronautics programs, it is necessary to continuously evaluate and improve the computational tools used for research and design at NASA. One such code is the Glenn-HT code which is used at NASA Glenn Research Center (GRC) for turbomachinery computations. Although the code has been thoroughly validated for turbine heat transfer computations, it has not been utilized for compressors. In this work, Glenn-HT was used to compute the flow in a transonic compressor and comparisons were made to experimental data. The results presented here are in good agreement with this data. Most of the measures of performance are well within the measurement uncertainties and the exit profiles of interest agree with the experimental measurements.

Introduction

NASA Rotor 37 test case, designed and tested originally by Reid and Moore (Ref. 1) at NASA Lewis, presently Glenn Research Center, is a low aspect ratio inlet stage for an eight-stage core compressor with a 20:1 pressure ratio (Ref. 2). It was retested at NASA Glenn in isolation to avoid the interaction effects. As such the test case is ideal for code verification. The specifications, given in Table I, are the intended design parameters and are reproduced from Suder (Ref. 2). Much data representing performance and flow field variables can be found in (Ref. 2) which has been used for CFD validation.

Given the steady nature of the flow and the isolation of the rotor blade row, the numerical simulation of this case should be rather straight forward. But, the results of the IGTI 1994 blind test case (unpublished) showed a large variation in the results and often large deviation from the experimental results. A new round of comparison with the data has been undertaken to assess the current state-of-the-art in some of the codes used at NASA. Glenn-HT although used mostly for heat transfer computations in turbine blades and passages was thus assessed by computing the case and comparing with the data.

The code Glenn-HT is a general structured multi-block code. Blocks are full face matching and can be arranged in an unstructured fashion. Alternatively, the computational domain can be formed from grid zones which are brought together at non-matching interfaces though this was not done in the present work. This makes the code ideal for complicated geometries and near wall solutions for which the near wall resolution is important. Glenn-HT uses a finite volume scheme. It uses a second order central difference scheme plus a fourth order artificial dissipation. A pressure switch turns on a second order dissipation that is activated by shocks. Thus near shock solutions are first order accurate. Glenn-HT code uses a multi-grid solver and utilizes a Runge-Kutta explicit smoother. Further acceleration is achieved with residual smoothing. A second order upwind method described by Huynh (Ref. 3) of NASA Glenn is also available. As for turbulence models, Glenn-HT uses Wilcox's $k-\omega$ as the default model (Ref. 4). The code also uses other turbulence models including the v^2-f model of Durbin (Refs. 5 and 6) and a Reynolds Stress model developed by Wilcox (Ref. 4). Glenn-HT has found use in external and internal flows including heat transfer on turbine blades including tip clearance (Refs. 7 and 8), film cooling flow and heat transfer and conjugate heat transfer (Ref. 9). Other applications have included internal heating of airplane wings (Ref. 10). As for unsteady applications, solutions have been obtained for external turbine flows and heat transfer (Refs. 11, 12, and 13) and external flows resulting from pulse detonation (unpublished). Glenn-HT code has never been used for compressor flows and the present application is new.

TABLE I.—ROTOR 37 DESIGN PARAMETERS

PARAMETER	DESIGN VALUE
Rotor Total Pressure Ratio	2.106
Rotor Total Temperature Ratio	1.270
Rotor Adiabatic Efficiency	0.877
Rotor Head Rise Coefficient	0.333
Flow Coefficient	0.453
Mass Flow, kg/s	20.188
Rotor Wheel Speed, rpm	17188.7
Rotor Tip Speed, m/s	454.14
Hub / Tip Radius Ratio	0.70
Rotor Aspect Ratio	1.19
Number of Rotor Blades	36
Blading Type	Multiple Circular Arc (MCA)

In an earlier incarnation, the code Glenn-HT was called TRAF3D. Arnone and Ameri using TRAF3D participated in the blind test case involving the present data (Ref. 14). The TRAF3D code, at the time, was a single block code that used a ‘C’ grid topology and employed an algebraic turbulence model. In the blind test case the tip clearance was modeled without gridding the gap. In the present work the tip gap was accounted for by meshing the tip clearance.

Results

In Figure 1, two views of the multi-block grid generated for the blade passage and the clearance is shown. The grid was generated using the GridPro commercial software. A close-up view of the tip grid is included that shows the refinement of the tip grid. The grid contains approximately 1.8 million cells. Only one grid resolution was tried but owing to the very large number of grid cells used in the simulations, no further refinement was done for this case. Due to the topology of the grid, the resolution of the grid on the tip of the blade, is as fine as the grid on the suction or the pressure side. Including the resolution of the grid on the casing, the tip region contains 49 grid points across the span of the tip clearance which was 0.4 cm as reported by Suder (Ref. 2).

Earlier studies of tip clearance effect (Refs. 15 and 16) have been made with far fewer points in the tip. The inlet and exit were placed so that they matched the locations recommended by the blind test case. The domain was subdivided into many blocks which was a result of grid generation. The blocks were formed into groups of approximately equal sizes in terms of grid points to take advantage of the multi-processing capability of the code.

For turbulence model, the Low Reynolds number $k-\omega$ model of Wilcox (Ref. 4) was used. Chima (Ref. 17) tested a newer version of the $k-\omega$ model (Ref. 18) and found it to perform better than the older version for the present case. We have not used any other models for this test case.

The dissipation is scaled with the grid resolution in Glenn-HT. Still only small amount of fourth order artificial dissipation was needed to achieve stable solution and convergence. The solution at 70 percent span section, in terms of Mach number for the near peak efficiency condition (discussed later), is shown in Figure 2. The bow shock and the shock/boundary layer interaction can be seen to have been simulated with good resolution. On the blade suction side, shown in Figure 3, the streamlines show a small recirculation zone that extends from the hub to the tip. This recirculation zone results from the interaction of the shock with the boundary layer as seen in Figure 2.

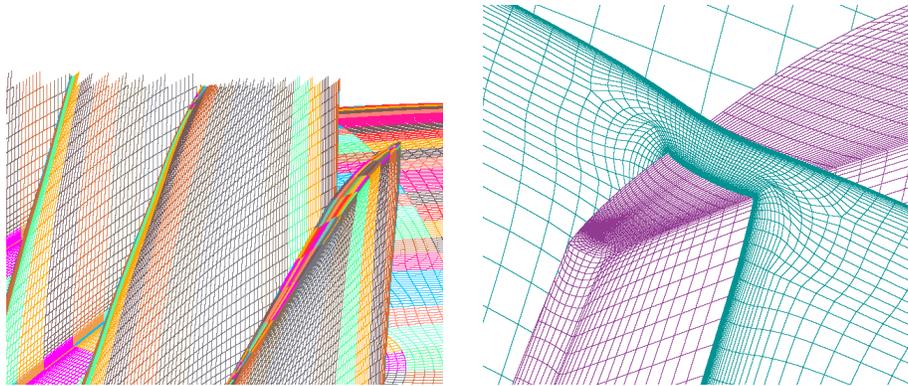


Figure 1.—Grid for the rotor 37 computations.

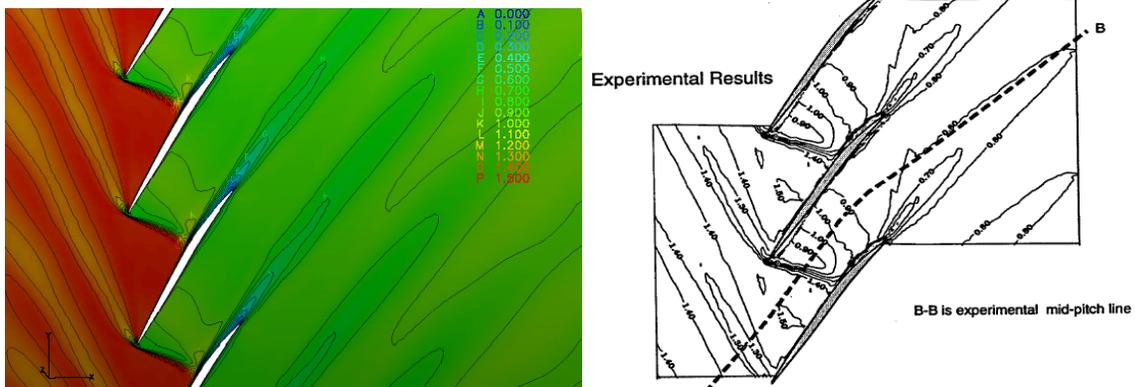


Figure 2.—Blade to blade Mach number contours, CFD (left) and experiment (right).

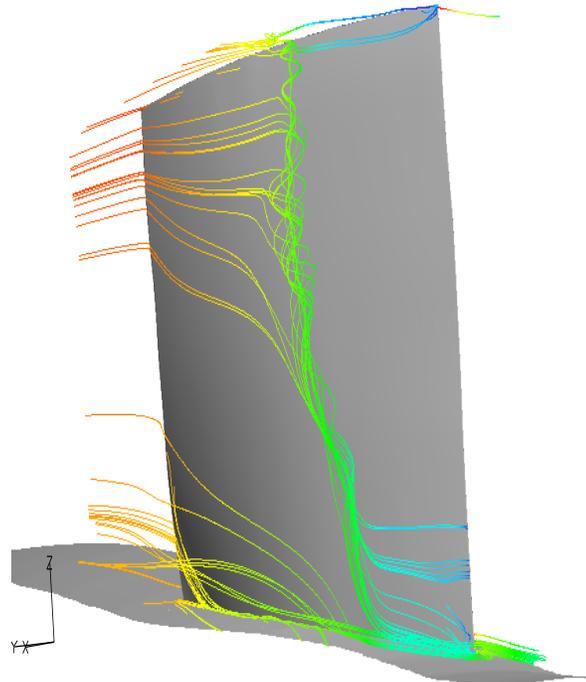


Figure 3.—Flow streamlines showing some important features of the flow.

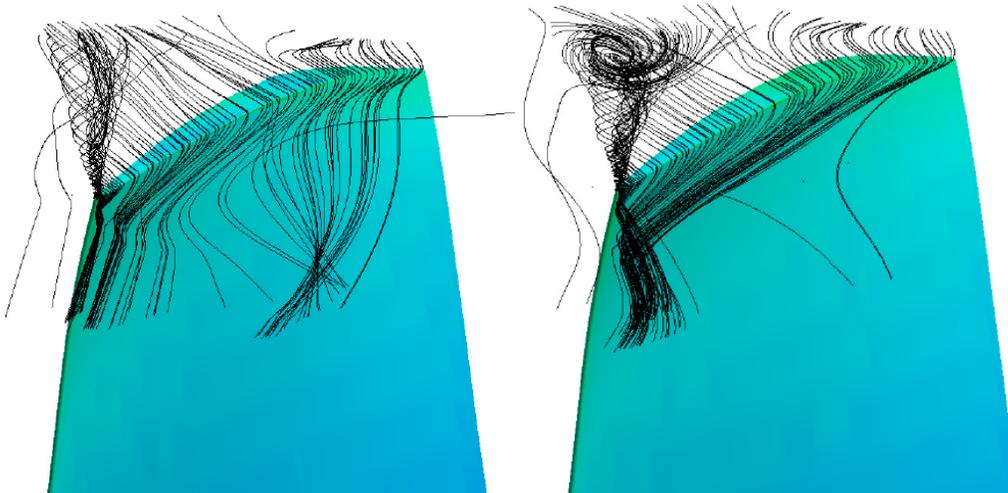


Figure 4.—Streamlines showing the flow over the blade tips at the near peak efficiency (left) and near stall (right) conditions.

Figure 4 shows the streamlines showing the flow over the blade tip for the near peak and near stall efficiency conditions. Near the leading edge the streamlines form into the tip vortex. As suggested by the streamlines, the tip vortex for the near peak condition is tighter and causes a smaller blockage to the flow than the near stall condition. Tip clearance flow has an important role to play in the performance of the rotor. Figure 5 shows the tip flow in the near leading edge region and in the mid-chord region. The flow recirculation appears to be larger in the near leading edge region and appears to diminish further down the blade. The resolution in the flow as can be seen is quite good. This is evidenced by the presence of core of the tip and edge vortices. The blade surface for the top two figures is colored according to pressure. The tip is seen to unload and assume the suction side pressure level very near the tip.

Performance Measures

One and a 10 percent boundary layer thickness profiles were prescribed at the inlet to the computational domain. The quantities of interest were found at the inlet and the exit of the passage at what was referred to as stations 1 at $x = 4$ cm upstream and station 4 at $x = 10$ cm downstream as given in (Ref. 2). Quantities were averaged at the measuring station.

Figure 6 shows the total pressure rise (outlet/inlet) versus the normalized mass flow through the passage. The map was formed by initially computing the choked mass flow for a static exit pressure ratio of 1.05. The static pressure ratio was subsequently raised in increments up to a value of 1.22 corresponding to the stall condition and the exit total pressure rise and mass flow rate noted. The results are plotted against the experimental data provided by Suder and Celestina (Ref. 12). The experimentally measured boundary layer thickness is 0.82 cm which is 10 percent of the inlet passage height. The experimental choked mass flow was 20.93 kg/sec. The choked mass flow rate computed for a 10 percent inlet boundary layer thickness was 20.56 kg/sec which is 1.76 percent below the experimental value. A 1 percent boundary layer thickness was attempted for which the computed choked mass flow was 20.85 kg/sec or 0.38 percent below the measured value. As shown, an increase in the inlet boundary layer thickness causes a reduction in the choked mass flow. The resulting map, in Figure 6, for the two boundary layer thicknesses produce the same results for pressure ratio, temperature ratio and adiabatic efficiency. We will present the results for the 10 percent boundary layer thickness as in the experiment, but we would note here that the remaining results for the two cases are quite similar.

Figure 6 shows the normalized total pressure, total temperature and adiabatic efficiency as a function of normalized mass flow rate. The agreement with the experimental data for the temperature ratio is quite good, but the computed pressure ratio and adiabatic efficiencies are low.

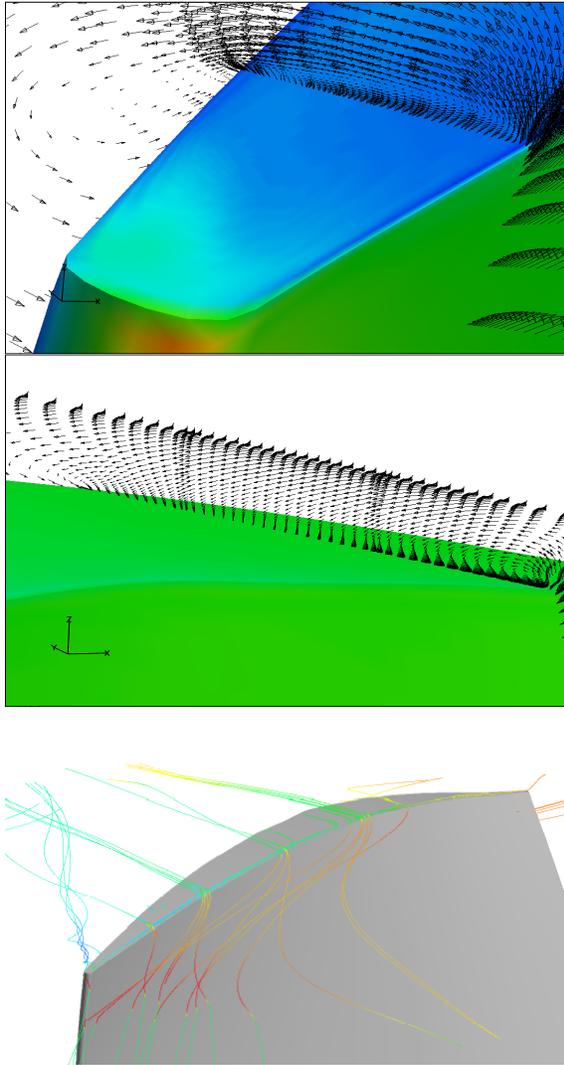


Figure 5.—Tip flow showing the recirculation zone in the tip clearance, top: near L.E. and middle in the mid chord region. The bottom figure shows the edge vortex along the blade tip.

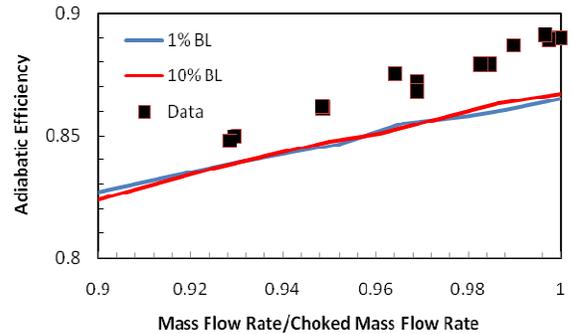
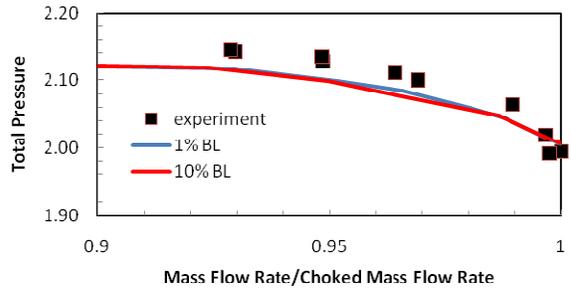
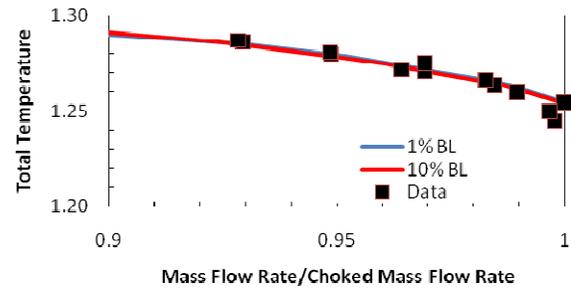


Figure 6.—Total pressure, total temperature and adiabatic efficiency versus normalized mass flow.

Exit Profiles

Two points on the map were chosen for further comparison. A near peak efficiency and a near stall point. The near peak efficiency was at a normalized mass flow rate of 0.98 and the near stall point was at a normalized mass flow rate of 0.925. We have chosen the pressure ratios of 1.12 and 1.22 as our corresponding points based on the mass flows and proximity to the runs we have made.

Figure 7 shows the spanwise variation of the tangentially averaged exit (downstream station $x = 10$ cm) total pressure, for the points on the map designated as near peak efficiency and near stall. The near hub deficit in the total pressure, evident in the data, has been captured. The results of the 1994 blind test case for the near peak efficiency results are also presented for comparison. For the total temperature, in Figure 8, the trends are captured and the agreement is quite good. The bands in which the results would be considered in good agreement with the data are shown for both total temperature and for the total pressure. The agreement is thus quite good for both of these measures. The large over-prediction in total

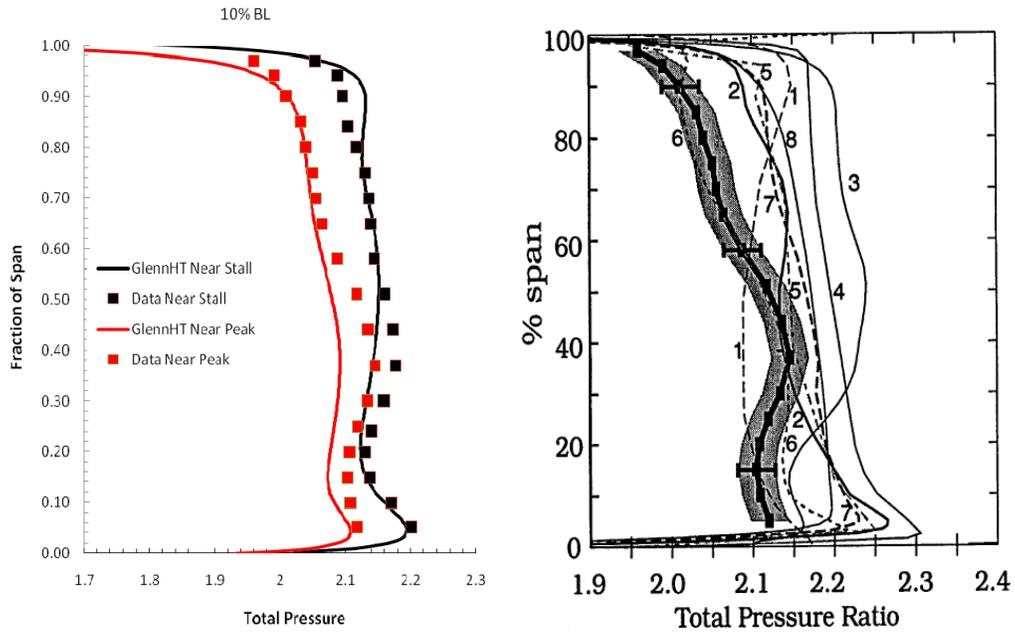


Figure 7.—Pitchwise average of total pressure across the span (left) and the results of 1994 blind test case.

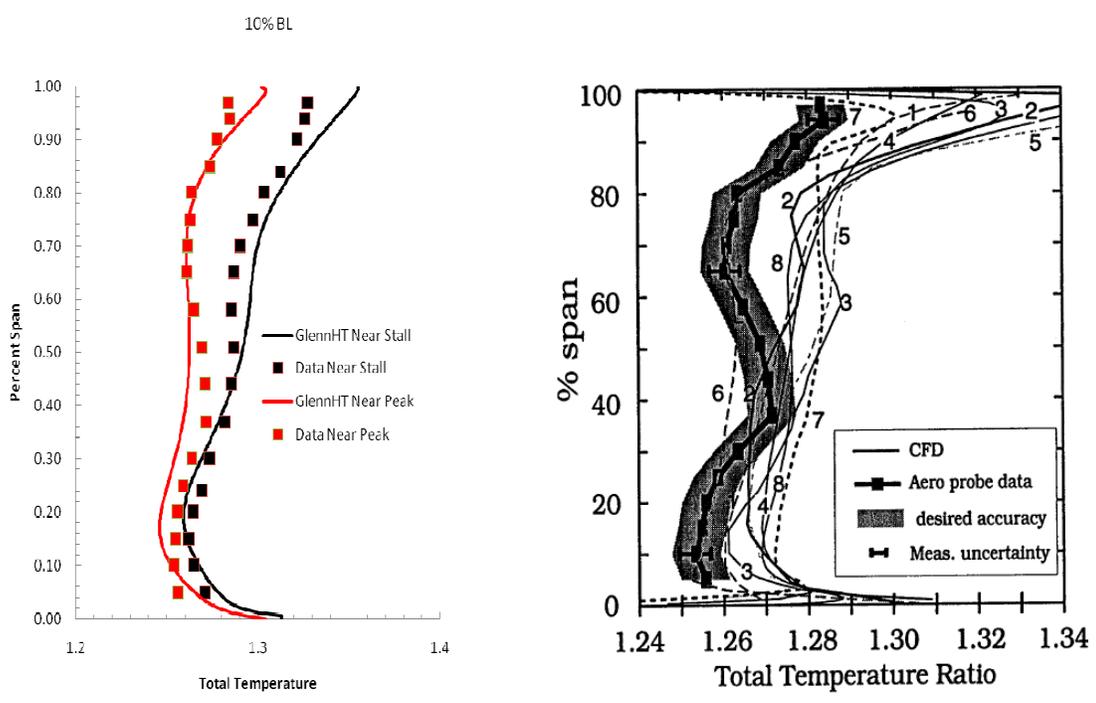


Figure 8.—Pitchwise average of total temperature across the span (left) and the results of 1994 blind test case.

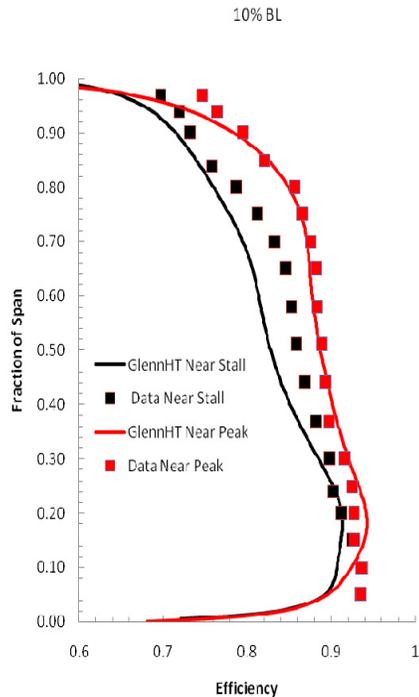


Figure 9.—Pitchwise average of adiabatic efficiency at the exit.

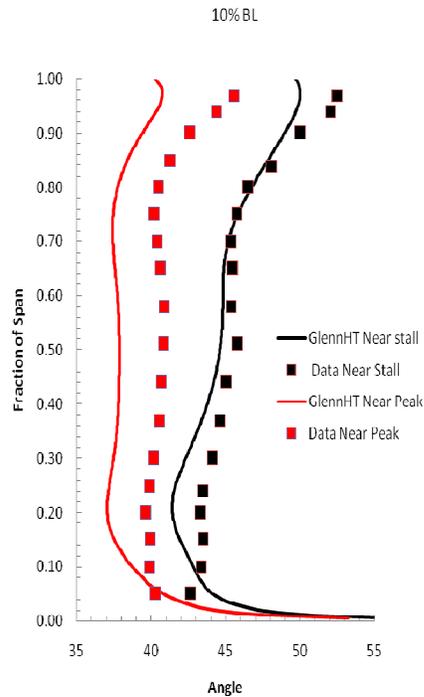


Figure 10.—Pitchwise average of exit angle at the exit.

temperature near the case that was discussed by Chima (Ref. 17) has been largely eliminated. This could be due to the fine grid resolution used in the blade tip region. The efficiency and the exit angle are shown in Figures 9 and 10. The agreement with the efficiency is quite good. We have not been able to reconcile the contradiction presented by the agreement of adiabatic efficiency with the data shown in Figure 9 and lack of agreement shown in Figure 6 in the map. Finally, the exit angle distribution is also shown in Figure 8. There is a maximum of 3° difference between the measured and computed exit angles.

Summary

The code Glenn-HT was used to simulate the flow in NASA rotor 37. A fine grid with a finely gridded tip clearance was employed. The default turbulence model in the code namely, the $k-\omega$ model was utilized. The agreement with the data at the downstream measuring station was quite good. The performance map was fairly well predicted. Glenn-HT is mainly used for thermal analysis in gas turbines. As a code, Glenn-HT is first and foremost a flow solver. Heat transfer capabilities are added features to enable thermal analysis the success of which depends on the underlying flow analysis scheme and the associated level of accuracy. From the results of this assessment, it can be concluded that Glenn-HT may be confidently used for compressor flows. It is expected that even better agreement with the data can be attained by using more appropriate turbulence models.

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